

# Simultaneous Vibration Reduction and Performance Enhancement in Rotorcraft Using Actively Controlled Flaps

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## Abstract

A computational study of helicopter vibration and rotor shaft power reduction is conducted using actively-controlled trailing-edge flaps (ACFs), implemented in single and dual flap configurations. Simultaneous vibration reduction and performance enhancement is demonstrated under level flight condition at high advance ratios, where dynamic stall effects are significant. Power reduction is achieved using the adaptive Higher Harmonic Control (HHC) algorithm in closed loop, with 2-5/rev flap control harmonics. This approach is compared with an off-line, nonlinear optimizer LSQNONLIN available in the MATLAB package, and favorable comparisons are obtained. Subsequently a parametric study of flap spanwise location is conducted to determine the optimal flap location for power reduction. Finally the effectiveness of ACF approach for power reduction is compared with that of the conventional individual blade control (IBC) approach. The simulation results clearly demonstrate the potential of the ACF system for power reduction as well as simultaneous vibration and power reduction.

<u>Nomenclature</u>		<b>T</b>	
$c$	Blade chord	$\mathbf{u}_k$	Sensitivity, transfer matrix between control inputs and objective function
$c_c$	Flap chord	$\mathbf{u}_{k,opt}$	Control input vector, $k^{\text{th}}$ control step
$C_T$	Rotor thrust coefficient		Optimum value of control input vector
<b>D</b>	Matrix defined to be $\mathbf{T}^T \mathbf{Q} \mathbf{T} + \mathbf{R}$	$x_c$	Spanwise location of center of control surface
$F_{HX4}, F_{HY4}, F_{HZ4}$	Nondimensional 4/rev hub shears	$\mathbf{z}_k$	Objective vector, $k^{\text{th}}$ control step
$M_{HX4}, M_{HY4}, M_{HZ4}$	Nondimensional 4/rev hub moments	$\alpha$	Rotor disk angle of attack
$M_{Hz1}$	Yawing moment about rotor hub	$\delta$	Flap deflection angle
$N_b$	Number of rotor blades	$\delta_{Nc}, \delta_{Ns}$	N/rev cosine and sine amplitude of $\delta$
$P_R$	Rotor shaft power	$\alpha_R$	Relaxation coefficient for control algorithm
$R$	Rotor blade radius	$\beta_p$	Blade precone angle
<b>Q</b>	Weighting matrix for objectives to be reduced	$\gamma$	Lock number
<b>R</b>	Weighting matrix on control input	$\mu$	Helicopter advance ratio
		$\omega_F, \omega_L, \omega_T$	Blade flap, lead-lag and torsional natural frequencies
		$\Omega$	Rotor angular speed
		$\psi$	Rotor azimuth angle
		$\theta_{tw}$	Built-in twist angle

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## Introduction and Background

Specifications for noise and vibration levels in rotorcraft are continuously increasing in stringency, thus motivating research related to active noise and vibration reduction. Desirable vibration levels have been identified to be below 0.05g to provide passengers with “jet smooth” ride. A number of active control techniques have emerged for effective vibration reduction [1, 2], as illustrated schematically in Fig. 1. These approaches generally fall into one of two categories: (a) active control approaches aimed at reducing vibrations in the rotor before they propagate into the fuselage, and (b) active control approaches implemented in the fuselage using an approach known as active control of structural response (ACSR). Within the first category of active control, where the primary objective is to reduce vibrations in the rotor, two approaches have emerged. These are (1) higher harmonic control (HHC) where the blades are activated in the non-rotating swashplate by introducing pitch commands, and (2) individual blade control (IBC) where each blade can be controlled independently in the rotating frame. Several implementations of IBC are available: (i) the conventional or earliest implementation based on pitch actuation at the blade root in the rotating system, (ii) actively controlled partial-span trailing-edge flaps, and (iii) the active-twist rotor where the entire blade is twisted by piezoelectric fiber embedded in the blade. Additional descriptions of these approaches can be found in Refs. 1 and 2.

Among these approaches the HHC and IBC were developed earlier and have been tested extensively in the wind tunnels as well as flight tests. Excellent vibration reduction of more than 80% has been demonstrated using these approaches. Subsequently, these approaches have also been considered for noise control, particularly in the blade vortex interaction (BVI) flight regime. Potential to reduce noise levels by 4-10dB under BVI conditions has been demonstrated in wind tunnel tests, with various helicopter configurations [3-5]. It is important to note that these active control approaches rely primarily on what is known as the conventional HHC algorithm in rotorcraft community for vibration reduction [1, 2, 6, 7].

More recently, actively controlled flaps (ACFs) have emerged as an efficient means of the active con-

trol of vibration due to BVI as well as the alleviation of dynamic stall induced vibrations [1, 8-16]. These studies on ACF system implemented in both single and dual flap configurations have demonstrated vibration reduction comparable to those achieved with HHC or conventional IBC, with no adverse effects on helicopter airworthiness and significantly less power requirement as compared to the blade root actuation approaches [1]. Wind tunnel tests have also shown the feasibility of the ACF for vibration reduction [16, 17]. During the Smart Material Actuated Rotor Technology (SMART) program conducted by Boeing [18, 19], a full scale piezoelectrically actuated flap system for vibration and noise control for a five-bladed bearingless MD-900 rotor has been tested on a whirl tower to demonstrate control effectiveness. In Europe, a full scale BK117 with three actively controlled, piezoelectrically actuated flaps has been flight tested by Eurocopter Germany in the open loop mode on September 6, 2005 and the flight test will continue into 2006 [20].

In Refs. 11-13, the effectiveness of ACF system to reduce the vibrations in the high advance ratio flight regime, where dynamic stall effects are known to be important, was studied by Depailler and Friedmann. The simulation indicated that ACF was successful in alleviating dynamic stall induced vibrations, thus demonstrating the capability of the ACF systems to reduce vibrations due to multiple sources. Recently, a comprehensive helicopter simulation code was developed using a unified approach for the prediction and active control of vibratory loads and blade-vortex interaction noise [21-25]. Considerable potential for active noise reduction and simultaneous vibration and noise reduction have been demonstrated using actively controlled flaps with 2-5/rev components on a rotor resembling the MBB BO-105 hingeless rotor [22, 23]. The capability of the ACF system has also been demonstrated using a bearingless rotor configuration resembling the MD-900 rotor in Ref. 24.

Despite their success for vibration and noise control demonstrated in experiments as well as numerical studies, these active control techniques are still mostly in preliminary flight test stages [26]. Concerns such as: cost, interference with the primary flight controls, and the power usage of the system have thus far prevented the practical implementation of such devices on a production helicopter. Therefore, it is important to study further the capability of such active control devices, so as to reap the largest potential benefit on the fairly sizeable cost associated with installing such active control

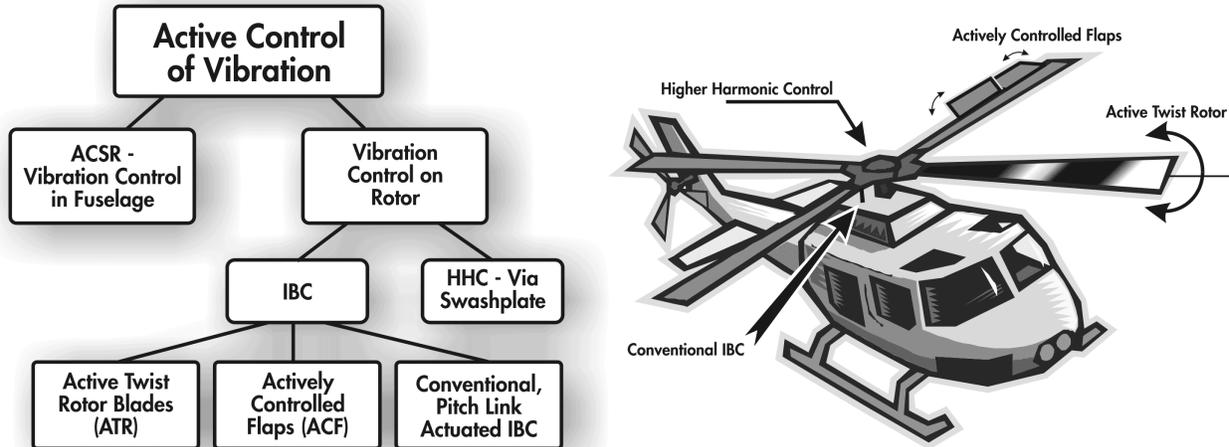


Figure 1: An Overview of Active Control Techniques.

systems in rotorcraft. One of the most desirable objectives is performance enhancement with the active device, or at least prevent undue performance penalty when deploying such a device for vibration or noise control.

A wind tunnel study by Shaw et. al. [27], which was intended primarily to demonstrate the effectiveness of HHC system for vibration reduction, also provided a preliminary assessment of the system for performance enhancement. The test was conducted on a scaled three-bladed CH-47D rotor, at two cruise airspeed of 135 knots and 160 knots ( $\mu \approx 0.30$  and  $0.35$ ), respectively. Pure  $2/\text{rev}$  HHC inputs with  $2^\circ$  amplitude were used for performance enhancement, and the optimal phase angle was determined experimentally. It was found that the power required in trim was reduced substantially by 6% at 135 knots and 4% at 160 knots. In another study, full-scale wind tunnel tests of a MBB BO-105 rotor were conducted at NASA Ames in the  $40 \times 80$  foot wind tunnel [4, 28]. An IBC system was tested in the open-loop for vibration and noise reduction as well as rotor power reduction. Rotor power reductions of up to 7% were demonstrated using  $2/\text{rev}$  IBC at advance ratios of 0.40 and 0.45, but no power reduction could be achieved at advance ratio of 0.30.

An analytical study conducted by Nguyen and Chopra [29] examined the effects of HHC on a scaled three-bladed rotor similar to the one tested in Ref. 27. A power reduction of 3.8% was obtained using  $2/\text{rev}$  HHC inputs with  $2^\circ$  amplitude. Cheng and Celi [30] performed a computational study of power reduction using  $2/\text{rev}$  HHC inputs. The rotor model

used was fairly simple, with rigid blade dynamics and a dynamic inflow model. The study noted that rotor power reductions were possible when using properly phased open-loop HHC input, at an advance ratio of  $\mu = 0.3$ . Reference 30 was based on table look-up aerodynamics and a nonlinear drag model; and it was emphasized that power reductions could only be obtained by simulation when a nonlinear drag model was used. A subsequent study by the same authors used numerical optimization techniques to determine the optimal  $2/\text{rev}$  input, and included a free wake model [31]. With the addition of free wake, the amount of power reduction that could be simulated was almost eliminated.

Performance enhancement using the ACF approach has only been attempted in a few studies thus far. A model rotor equipped with cam operated trailing edge flaps was tested in wind tunnel by Straub [32]. Effect on rotor performance with  $2/\text{rev}$  flap actuation was evaluated at advance ratios of 0.25 and 0.30, but the results were considered inconclusive due to issues associated with the accuracy of the measurements obtained during the test. A preliminary computational study on rotor power reduction for a rotor resembling the MBB BO-105 was conducted in Ref. 25. In this study a single flap configuration with deflections limited to  $4^\circ$  degrees was examined in the open loop mode, under BVI conditions at  $\mu = 0.15$ . It was found that power reduction could only be obtained when the torsional fundamental frequency of the rotor was reduced from the value of the MBB BO-105 to a lower value of 2.5.

When using an ACF for vibration or noise reduc-

tion the deflections of the flap increase the drag of the blade and therefore vibration and noise reduction are accompanied by a performance penalty. The fundamental question is therefore whether it is possible to use the ACF system without incurring a power penalty. Motivated by this desire the overall objective of the paper is to examine simultaneous vibration reduction and performance enhancement using the ACF based approach. Rotor power reduction using ACF approach, employing the HHC algorithm in the closed loop, will be emphasized. Based on the concise review of previous research provided in this introduction on rotor power reduction, the rotor power reduction problem at high advance ratio ( $\mu = 0.35$ ) will be considered. While the noise aspect associated with this problem is also of interest, the noise emissions at such advance ratios are typically dominated by high speed impulsive (HSI) noise [33] and this problem is computationally very intensive and therefore will be considered in follow-on research. The specific objectives of the paper are provided next:

- assess the potential for rotor power reduction using the ACF system by applying the HHC control algorithm;
- examine the mutual interaction between power reduction and vibration reduction;
- explore the potential of simultaneous power and vibration reduction using the ACF system;
- determine the sensitivity of power reduction to flap spanwise locations so as to optimize flap performance;
- compare the effectiveness for power reduction using the ACF approach with conventional IBC approach;
- conduct a study of power reduction, vibration reduction as well as simultaneous reduction using off-line nonlinear optimizers, and compare the results with those obtained using the HHC algorithm.

Achieving reduced vibration or noise levels without undue performance penalty is central to practical implementation of the ACF system; and it has a key role governing the feasibility of implementing an ACF system in a practical setting.

## Mathematical Model

The present study is based on a comprehensive rotorcraft aeroelastic analysis tool that accounts for the effects of dynamic stall at high advance ratios, as described in detail in Refs. 11–13. The power reduction studies conducted in the present research will be carried out at a high advance ratio of 0.35, where the dynamic stall effects are important. The fundamental ingredients of the aeroelastic model are concisely summarized in the following subsections.

### Structural Dynamic Model

The structural dynamic model consists of an isotropic hingeless rotor blade, which is cantilevered at the hub and has fully coupled flap-lag-torsional dynamics including nonlinearities due to moderate blade deflections [8]. The aeroelastic model is capable of simulating rotors with single or dual actively controlled partial span trailing edge flaps mounted on the rotating blade as depicted in Fig. 1. The equations of motion are discretized using the global Galerkin method, based upon the free vibration modes of the rotating blade. Three flapping modes, two lead-lag modes and two torsional modes are used when computing the numerical results given in the results section.

### Aerodynamic Model

Blade section aerodynamic loads for *attached flow* are calculated using a rational function approximation (RFA) approach as described in Ref. 9. The RFA approach is a two-dimensional, unsteady time-domain aerodynamic theory that accounts for compressibility, variations in the oncoming velocity and a blade-flap configuration. Aerodynamic cross-sectional loads including lift, moment and flap-hinge moment can be calculated, along with chordwise pressure distribution [21] which is required in acoustic calculations. The RFA model for the blade-flap combination is linked to a free wake model [10], which produces non-uniform inflow distribution. For the *separated flow* regime, unsteady aerodynamic loads are calculated using the ONERA dynamic stall model described by Petot [34]. The augmented aerodynamic states associated with RFA attached flow states and ONERA separated flow states are combined to produce the time-domain, state space aerodynamic model. Furthermore, a simple, linear drag model that accounts for additional drag

due to flap deflection, that increases the power required for the rotor to operate, is also implemented [12].

### Solution Procedure

The combined structural and aerodynamic equations form a system of coupled differential equations that can be cast in state variable form. They are then integrated in the time domain using the Adams-Bashforth DE/STEP predictor-corrector algorithm. A propulsive trim procedure is implemented in level flight condition; where six equilibrium equations (three forces and three moments) are enforced. The trim equations are solved in a coupled manner with the aeroelastic equations of motion. Hub vibratory loads are obtained by integrating the distributed aerodynamic and inertial loads over the blades. Average rotor power is defined as the instantaneous power required to drive the rotor at a constant angular velocity  $\Omega$  averaged over one revolution,

$$P_R = \frac{\Omega}{2\pi} \int_0^{2\pi} -M_{Hz1}(\psi) d\psi, \quad (1)$$

where  $M_{Hz1}$  is the total yawing moment about the hub and includes the effect of unsteadiness, compressibility, dynamic stall (if applicable), and the additional drag due to flap deflection. The negative sign in front of  $M_{Hz1}(\psi)$  is due to the fact that it represents the torque about the rotor shaft due to the loading on the blades, and therefore the engine must supply a torque equal to  $-M_{Hz1}(\psi)$  to maintain a constant angular velocity [8]. Equation 1 is a general expression valid for blades with or without actively controlled flaps.

### Control Strategies

The Higher Harmonic Control (HHC) algorithm [6, 7, 14] has been used successfully for both vibration and noise reduction; as well as simultaneous vibration and noise reduction. Furthermore, several variants of this algorithm, including a relaxed and an adaptive version, have been shown to improve the robustness of the algorithm [7]. An adaptive form of this algorithm was successfully applied in the closed loop noise control as well as simultaneous vibration and noise control in Ref. 22. In the present study, the adaptive HHC algorithm is used for rotor power reduction, where the objective

function consists of averaged rotor power over one revolution. For simultaneous reduction of vibration and power, a combined objective which consists of vibration and power components is used, and an appropriate weighting matrix is used to adjust the control effort so as to achieve a desirable balance between vibration and power objectives. For the practical implementation of the algorithm, an appropriate control input weighting is chosen such that the maximum flap deflection does not exceed  $4^\circ$ .

The HHC algorithm has been the subject of a recent paper [7], wherein the stability, robustness, and convergence properties of the algorithm together with a number of variants are addressed in detail. This algorithm is based on a linear, quasi-static, frequency domain representation of helicopter response to control inputs. The inputs to the algorithm are comprised of a combination of flap deflection harmonics with discrete frequencies of  $N_{\min}$ - $N_{\max}$ /rev. The total flap deflection is given as

$$\delta(\psi) = \sum_{N=N_{\min}}^{N_{\max}} [\delta_{Nc} \cos(N\psi) + \delta_{Ns} \sin(N\psi)] \quad (2)$$

For a four-bladed rotor, the flap deflection harmonics typically consist of 2-5/rev components for vibration reduction, as well as in noise control [22]. Most studies thus far have only examined the effects of 2/rev component for power reduction using open loop [27, 30, 31]; however, in the present study this limitation is removed and the whole range of harmonics from 2-5/rev will be used since the feedback controller based on the HHC algorithm is capable of choosing the most effective harmonics for the purpose of power reduction. These pitch deflections are related to the vibration or noise level magnitudes through a transfer matrix  $\mathbf{T}$ , given by

$$\mathbf{T} = \frac{\partial \mathbf{z}_k}{\partial \mathbf{u}_k}. \quad (3)$$

The control strategy is based on the minimization of a performance index that was originally developed for vibration reduction [6] which is a quadratic function of the quantities that are being reduced (vibration or noise)  $\mathbf{z}_k$  and control input amplitudes  $\mathbf{u}_k$ :

$$J(\mathbf{z}_k, \mathbf{u}_k) = \mathbf{z}_k^T \mathbf{Q} \mathbf{z}_k + \mathbf{u}_k^T \mathbf{R} \mathbf{u}_k, \quad (4)$$

The subscript  $k$  refers to the  $k^{\text{th}}$  control step, reflecting the discrete-time nature of the control. The time interval between each control step must be sufficient to allow the system to return to the steady state, typically in 3-5 revolutions, so that the vi-

bration or power levels can be accurately measured. The optimal control law is given by:

$$\mathbf{u}_{k,\text{opt}} = -\mathbf{D}^{-1}\mathbf{T}^T\mathbf{Q}\{\mathbf{z}_0 - \mathbf{T}\mathbf{u}_0\} \quad (5)$$

where

$$\mathbf{D} = \mathbf{T}^T\mathbf{Q}\mathbf{T} + \mathbf{R} \quad (6)$$

For a well-identified linear system the algorithm converges to the optimum value in a single step [7]. However, if the helicopter cannot be perfectly represented by a linear model, the optimal value will not be reached after the first step. The convergence issue of the algorithm was addressed in Ref. 7, where a relaxed version of the algorithm, described concisely below, was developed. Traditionally, the control input update is represented in iterative form as shown in Eq. (7):

$$\mathbf{u}_{k+1} = \mathbf{u}_k + \Delta\mathbf{u}_k. \quad (7)$$

In the relaxed variant of the algorithm, a relaxation factor  $\alpha_R$  is introduced,

$$\mathbf{u}_{k+1} = \mathbf{u}_k + \alpha_R\Delta\mathbf{u}_k, \quad (8)$$

where  $0 < \alpha_R < 1$ . This has been shown to increase the robustness of the algorithm at the expense of convergence speed [7]. An adaptive version [6, 7] of the HHC algorithm was also shown to be useful in the noise reduction studies in the presence of stronger nonlinearities, therefore it would also be applied in this study for power control. In the adaptive variant, the transfer matrix  $\mathbf{T}$  is identified online, using a recursive least-squares technique, following the method described in Ref. 7.

For vibration reduction (VR) studies, the vector  $\mathbf{z}_k$  consists of 4/rev vibration levels as represented by hub shears and moments, given in Eq. (9),

$$\mathbf{z}_{k,\text{VR}} = \begin{bmatrix} F_{HX4} \\ F_{HY4} \\ F_{HZ4} \\ M_{HX4} \\ M_{HY4} \\ M_{HZ4} \end{bmatrix} \quad (9)$$

When the controller is used for power reduction the objective vector  $\mathbf{z}_k$  is simply averaged rotor shaft power as given in Eq. (1).

$$\mathbf{z}_{k,\text{PR}} = [ P_R ] \quad (10)$$

For simultaneous reduction (SR) problems, a com-

bined output vector is defined

$$\mathbf{z}_{k,\text{SR}} = \begin{bmatrix} \mathbf{z}_{k,\text{VR}} \\ \mathbf{z}_{k,\text{PR}} \end{bmatrix}. \quad (11)$$

The weighting matrix  $\mathbf{Q}$  is used to adjust the control effort so as to achieve a desirable balance between the vibration levels and power reductions.

In order to evaluate the effectiveness of the HHC algorithm under the high speed forward flight conditions, where strong nonlinearities are present characterized by the onset of dynamic stall, the aeroelastic simulation code is coupled with MATLAB so as to utilize the built-in nonlinear optimization solvers to find potentially lower controlled values using the same active flap configurations. The nonlinear least squares optimizer LSQNONLIN is chosen for this purpose, which is a subspace trust region method and is based on the interior-reflective Newton method described in Refs. 35 and 36. This approach is intended for offline identification of best possible vibration and rotor power reduction, using the ACF approach. The results obtained using the online adaptive HHC algorithm, which is computationally much more efficient, are compared with those obtained with LSQNONLIN to determine the performance of the algorithm. The results obtained with the offline optimizer represent the best possible performance enhancement that can be achieved.

## Results

The results presented in this section were obtained for a helicopter configuration resembling a full-scale MBB BO-105 helicopter with a four-bladed hingeless rotor system. The properties of the helicopter configuration used in the computations are summarized in Table 1. The characteristics of the actively controlled flap configurations are given in Table 2, including a single servo flap configuration and a dual servo flap configuration. The helicopter is in level flight condition at a relatively high advance ratio of  $\mu = 0.35$ , and propulsive trim is used to trim the rotor.

### Rotor Power Reduction

In this section the potential of the ACF approach for rotor power reduction is explored. First the adaptive higher harmonic control (HHC) algorithm is used for rotor power control. The effect of power

Table 3: Summary of rotor power reduction during Power Reduction (PR) and Simultaneous Reduction(SR).

Objective	Power Reduction(PR)			Simultaneous Reduction(SR)		
	HHC		LSQNONLIN	HHC		LSQNONLIN
Controller	HHC		LSQNONLIN	HHC		LSQNONLIN
Flap Config.	Single	Dual	Single	Single	Dual	Single
Baseline Power	0.00670286	0.00681677	0.00670286	0.00670286	0.00681677	0.00670286
Controlled	0.00658692	0.00669656	0.00654460	0.00667611	0.00677104	0.00663603
Reduction(%)	1.73	1.76	2.36	0.40	0.67	1.00

Table 1: MBB BO-105 hingeless blade configuration.

Rotor Data	
$N_b = 4$	$c = 0.05498L_b$
$\omega_F = 1.12, 3.41, 7.62$	$C_{do} = 0.01$
$\omega_L = 0.73, 4.46$	$C_{mo} = 0.0$
$\omega_T = 3.17$	$a_o = 2\pi$
$\theta_{tw} = -8^\circ$	$\theta_{FP} = 6.5^\circ$
$\gamma = 5.5$	$\sigma = 0.07$
$\beta_p = 2.5^\circ$	
Helicopter Data	
$C_T/\sigma = 0.0714$	$\mu = 0.35$
$L_b = 4.91$ m	$\Omega = 425$ RPM

Table 2: Flap configuration.

$c_c = 0.25c$	
Single Servo Flap	
$x_c = 0.75L_b$	$L_c = 0.12L_b$
Dual Servo Flap	
$x_c^1 = 0.72L_b$	$L_c^1 = 0.06L_b$
$x_c^2 = 0.92L_b$	$L_c^2 = 0.06L_b$

reduction on vibratory loads is also examined during power reduction. The results for rotor power reduction are summarized in Table 3.

As shown in Table 3, the rotor power can be reduced by 1.73% and 1.76% compared to the baseline, using the single and dual flap configurations, respectively. This reduction is achieved in the presence of dynamic stall, which is important under the simulated flight condition at  $\mu = 0.35$ . Note that the baseline power is slightly higher for the dual flap case, under the same rotor trim settings. During the power reduction the maximum flap deflections required by the controller are less than  $3^\circ$ , as shown in Fig. 2. Furthermore, it can be seen that the flap deflections of the two flaps are similar in the dual flap configuration, which also resemble the flap deflection for the single flap case.

Figure 3 illustrates that the vibratory loads dur-

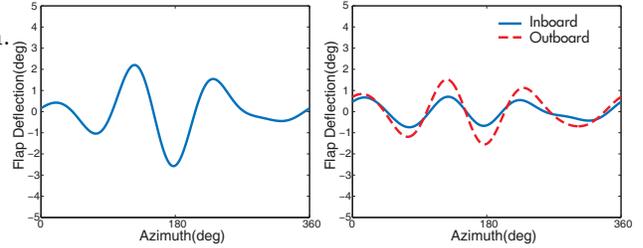


Figure 2: Flap deflection during power reduction.

ing power reduction are severely increased; by more than 100%, for both the single and dual flap configurations. This suggests that using a single objective of rotor power in the HHC controller could result in unacceptable vibration levels. Therefore, a combined objective which accounts for both vibration and power is required, such an approach is considered next.

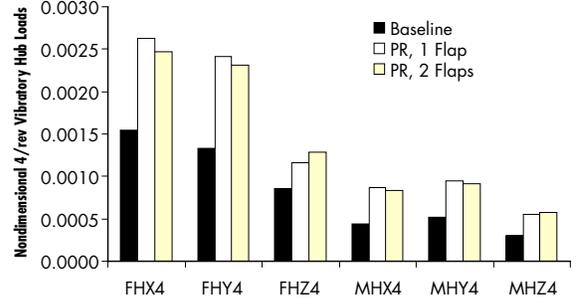


Figure 3: Vibration levels during power reduction.

### Simultaneous Vibration and Rotor Power Reduction

In this section, a composite objective function which combines vibratory loads and rotor power components is used with the HHC algorithm, so as to achieve simultaneous reduction of both vibration

and power. The results shown here are obtained using one weighting matrix: the vibratory hub shear components are weighted by a factor of 1, while the vibratory hub moments and rotor power components are weighted by a factor of 10. The rotor and flap configurations as well as flight conditions are identical to those used in the previous section.

Table 3 shows rotor power reduction of about 0.4% and 0.67% for the simultaneous control, using the single and dual flap configurations, respectively. Obviously the amount of power reduction that can be achieved for the combined objective is smaller than that has been demonstrated with the controller tuned for power reduction alone. The maximum flap deflections for these cases are less than  $3^\circ$ , as shown in Fig. 4.

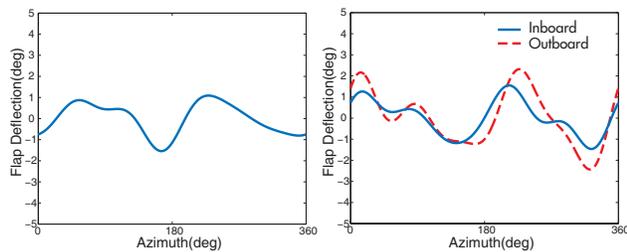


Figure 4: Flap deflection during simultaneous reduction.

The reduction of the vibratory loads during power reduction is shown in Fig. 5. For the single flap case, the individual hub shear and moment components are reduced between 28-78%, while the vibration objective is reduced by 68%. The combined vibration and power objective is reduced by 3.6%. Despite a slightly better power reduction during vibration suppression, the dual flap configuration produces slightly less vibration reduction compared to the single flap case. From these results, it can be seen that the ACF is capable of producing significant vibration reduction along with a small amount of power reduction, using small flap deflections of less than  $3^\circ$ .

### Vibration and Rotor Power Reduction Using Off-line Optimizer LSQNONLIN

In this section the results obtained using the nonlinear optimizer LSQNONLIN available in the MATLAB package are shown for vibration and power reduction. These results are also compared to those obtained previously using the adaptive HHC algorithm, to determine its effectiveness. In the optimization of

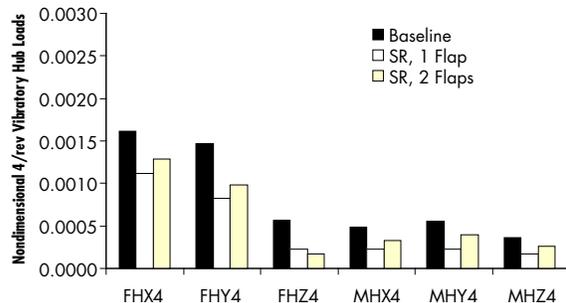


Figure 5: Vibration levels during simultaneous vibration and power reduction.

a nonlinear objective function local minima for vibration or power can be obtained during the search, therefore different flap inputs are applied as initial conditions. Ten initial flap inputs are generated randomly with the constraint that the maximum flap deflections not exceed  $4^\circ$ . The best optimization results using LSQNONLIN are chosen among these ten flap initial inputs along with the case with zero flap initial deflection.

The nonlinear optimizer LSQNONLIN is capable of achieving rotor power reduction of 2.36%, as shown in Table 3. This is superior to that achieved using the adaptive HHC algorithm; however, the performance of the adaptive HHC algorithm is reasonably good in addition to its numerical efficiency. The flap deflections required by the optimizer are shown in Fig. 6, where the maximum flap deflection is limited to be less than  $4^\circ$ . The vibration levels during power reduction are also increased significantly, shown in Fig. 7, which is similar to the results shown earlier using the HHC algorithm.

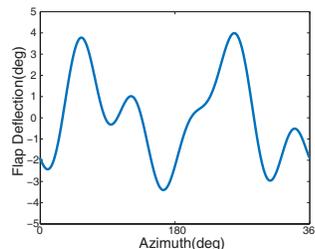


Figure 6: Flap deflection during power reduction using nonlinear optimizer LSQNONLIN.

For simultaneous vibration and power reduction, the rotor power is reduced by 1.0% as shown in Table 3 and the vibration objective is reduced by 55% as shown in Fig. 9. The flap deflections are shown in Fig. 8 and it is evident that flap deflections are less

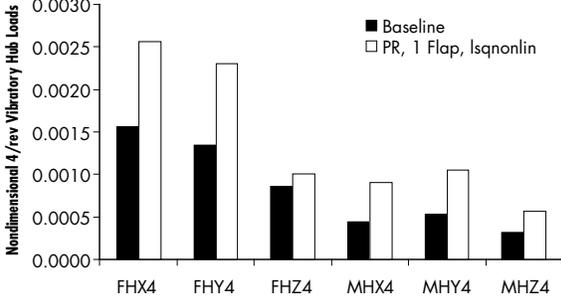


Figure 7: Vibration levels during power reduction using nonlinear optimizer LSQNONLIN.

than  $4^\circ$ . It should be noted that a different weighting matrix is used here, which puts more emphasis on power objective compared to the weighting matrix used in the HHC algorithm shown earlier. As a result, the degree of power reduction using LSQNONLIN is larger than that obtained using the HHC algorithm, whereas the degree of vibration reduction is slightly less than that found with the HHC algorithm.

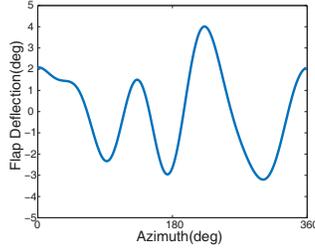


Figure 8: Flap deflection during simultaneous reduction using nonlinear optimizer LSQNONLIN.

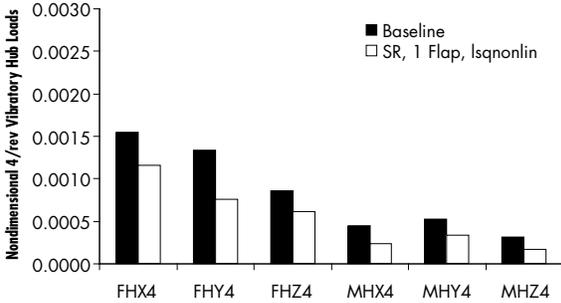


Figure 9: Vibration levels during simultaneous power and vibration reduction using nonlinear optimizer LSQNONLIN.

## Comparison of HHC to LSQNONLIN

A plot of the vibration objective  $J_{\text{vib}}$  versus the power objective  $J_{\text{power}}$  is depicted in Fig. 10. The figure shows the tradeoff between the competing objectives at all control steps for both LSQNONLIN and HHC algorithms. The blue circles represent the values of the objectives at all function evaluations during the course of optimization using LSQNONLIN. The boundary of these objectives, represented by the red line, shows an approximate Pareto-optimal curve illustrating the best tradeoff between the two competing objectives. The grey, black and green lines represent the power reduction, vibration reduction and simultaneous reduction using the HHC algorithm from the same baseline, respectively. It is quite interesting to note that the optimal vibration reduction that can be obtained using HHC reaches the Pareto optimal curve obtained from LSQNONLIN. Moreover, the best simultaneous vibration and power reduction can be achieved with HHC also approaches the optimal trade-off curve. However, it is evident that significantly better power reduction is obtained using LSQNONLIN, as shown in Fig. 10. Overall, the performance of HHC algorithm is quite good considering its numerical efficiency.

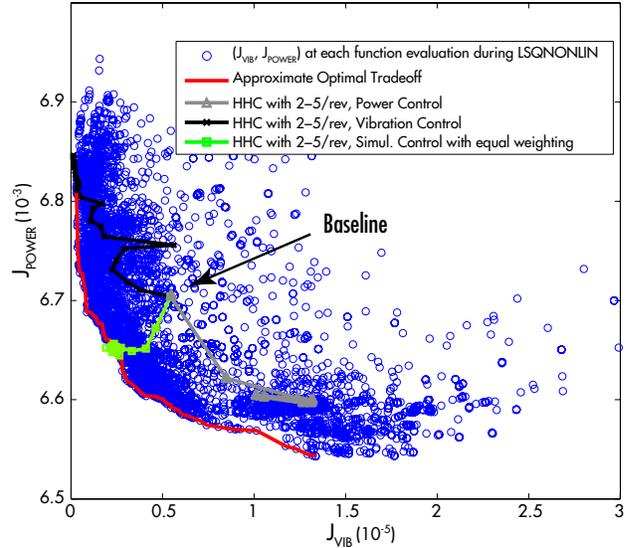


Figure 10: Comparison of optimization history using HHC and LSQNONLIN, showing  $J_{\text{power}}$  vs.  $J_{\text{vib}}$ .

## Effect of Flap Spanwise Locations

The flap configurations used in the previous sections, as shown in Table 2, are optimized for vibration reduction. Therefore, it is relevant to consider alternative flap spanwise locations which emphasize optimizing power reduction. The effect of different spanwise flap locations on the power reduction capability is considered next. This is accomplished by varying the location where a single servo flap is centered. The flap has a span equal to 12% of rotor radius and its chord is 25% of the blade chord, which is similar to the single flap configuration in Table 2. Three spanwise locations are considered, as shown in Table 4, where the flap with  $x_c = 0.94R$  is located at the tip of the blade.

It is interesting that the baseline rotor power is reduced when the flap is moved outboard. The rotor power reduction is somewhat less for the flap centered at  $0.85R$  than for one centered at  $0.75R$  in terms of power reduction percentage. When the flap is located at the blade tip the power reduction is greatest, almost 4%. However, the optimal flap deflection also alters significantly the rotor trim conditions, in particular rotor thrust. When comparing this characteristic to the flap placed at the other two locations, it is evident that for the inboard location the influence on the rotor trim is negligible. After retrimming the rotor while keeping this optimal flap deflection for power control, the degree of power reduction that can be achieved is reduced to 2%. Therefore, one has to be careful to ensure the similar rotor operating conditions are enforced when conducting power control studies.

## Comparison of ACF to Conventional IBC for Power Reduction

Power reduction using the conventional IBC approach is also considered here, so as to compare its effectiveness to the ACF approach. The controller used in the conventional IBC approach is also the adaptive HHC algorithm described earlier. The IBC control inputs also consist of a combination of 2-5/rev harmonics, similar to that used in the ACF study. Furthermore, the maximum IBC amplitude is restricted to be less than  $1^\circ$ .

Using the conventional IBC approach, the rotor power consumption is reduced to 0.00660756 from the baseline value of 0.00670286, representing a 1.4% reduction. This is similar to the reduction achieved using the ACF approach. The vibration

levels are significantly increased during power control using IBC as shown in Fig. 11, a phenomenon that was also observed during power reduction studies conducted using the ACF. The IBC time histories for power reduction are shown in Fig. 12.

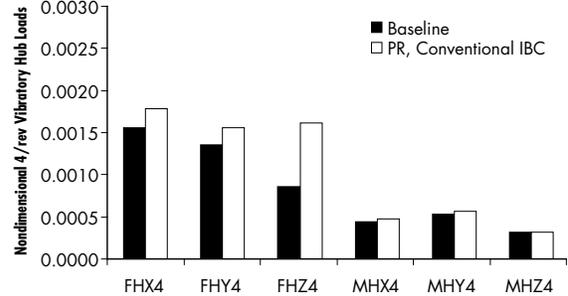


Figure 11: Vibration levels during power reduction using conventional IBC.

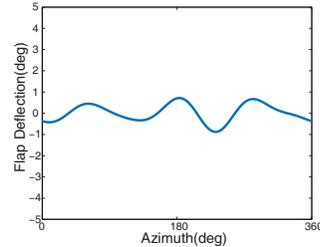


Figure 12: Flap deflection during power reduction using conventional IBC.

## Conclusions

The results presented in this paper have demonstrated that the ACF system implemented in a single or dual flap configuration is capable of simultaneously reducing vibration and enhancing rotor performance. The numerical simulations are obtained at a relatively high advance ratio of  $\mu = 0.35$  where the dynamic stall effects are significant.

The primary conclusions obtained in the course of this study are summarized below:

1. The ACF is an effective device for rotor performance enhancement, where a nearly 2% power reduction is achieved using a rotor configuration that resembles the MBB BO-105 under the simulated flight conditions.
2. The rotor power reduction can be implemented by using the adaptive HHC algorithm which

Table 4: Rotor power reduction with single flap located at various spanwise locations.

Flap Center ( $x_c$ )	0.75R	0.85R	0.94R
Baseline Power	0.00670286	0.00663195	0.00571054
Controlled	0.00658692	0.00654061	0.00548369
Reduction(%)	1.73	1.38	3.97
Power After Retrim	—	—	0.00559408
Reduction(%)	—	—	2.04

has been applied successfully for vibration and noise control in previous studies.

3. The rotor power reduction is accompanied by a significant increase in vibration levels, if the controller is tuned only for power enhancement.
4. Simultaneous reduction of vibration and rotor power reduction is feasible, and significant vibration reduction of 68% is achieved combined with a relatively modest amount of power reduction of approximately 1%.
5. A nonlinear optimizer LSQNONLIN available in the MATLAB package is also explored for off-line identification of best possible vibration and power reduction. Compared to LSQNONLIN, the adaptive HHC algorithm provides satisfactory performance with superior numerical efficiency, although considerably higher level of power reduction can be achieved using LSQNONLIN.
6. A parametric study of the flap spanwise location indicates that the flap placement near the blade tip can significantly improve power enhancement. However, the rotor trim is affected when the flap is placed near the tip, which requires retrimming the rotor.
7. The ACF system shows power enhancement capability that is comparable to the power enhancement that can be obtained with the conventional IBC approach.

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